

# Genesis Aero

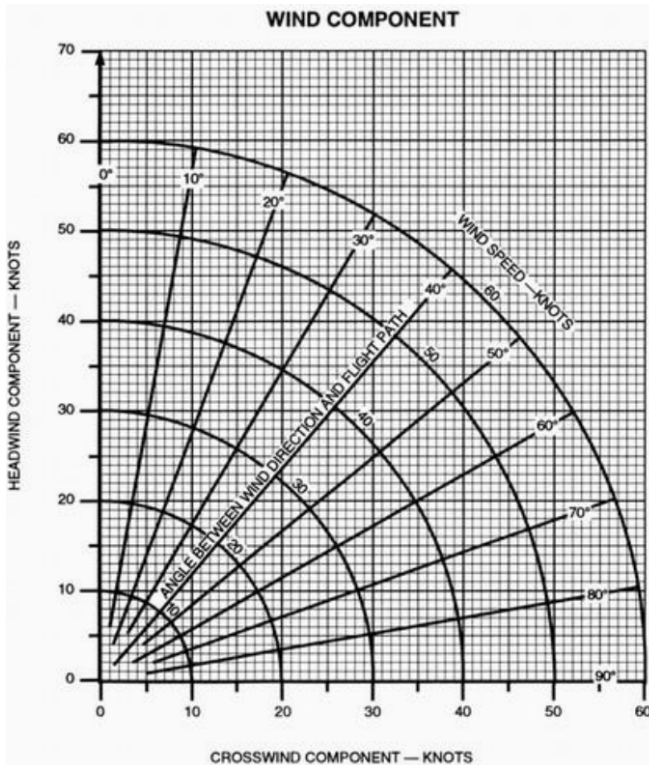
## Aircraft Weight and Balance

### Weight X Arm = Moment

	WT	ARM	MOMENT
Basic Empty Weight			
Pilot			
Front Passenger			
Rear Passengers			
Baggage (area 1)			
Baggage (area 2)			
Zero Fuel Weight/CG			
Fuel (6 lb/gal)			
<b>Ramp Weight</b>			
Minus fuel for Engine Start, Taxi, and Run-up			
<b>Takeoff Weight</b>			
Minus est. fuel burn (climb and cruise) 6lb/gal			
<b>Landing Weight</b>			

Pressure Altitude:	_____
Temperature:	_____
Density Altitude:	_____
Take-off Ground Roll:	_____
Total dist. to clear 50ft obst.	_____
Landing Ground Roll:	_____
Total dist. to land over 50ft obst.	_____
Headwind:	_____
Crosswind:	_____
Maneuvering Speed (Takeoff):	_____
Maneuvering Speed (Landing):	_____

Formulas:  
 PA = [29.92 - altimeter setting] x 1000 + field elevation  
 DA = (current temp - standard temp at field elev) x 120 + PA  
 Va = Va x  $\sqrt{\text{current weight} / \text{max gross weight}}$



Aircraft	Empty Wt	Moment
<b>N18LW</b>	1655.50	64,516.00
<b>N62AW</b>	1692.70	66,271.04
<b>N716MW</b>	1655.40	65,087.18
<b>N1232G</b>	1699.40	68,741.16
<b>N5327G</b>	1698.10	69,847.19
<b>N685SP</b>	1646.90	63,341.40
<b>N384CP</b>	2185.22	306,691.45
<b>N50MF</b>	2173.00	307,371.00
<b>N344DG</b>	2169.00	307,364.00
<b>N955W</b>	2328.00	324,615.00

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